



US Department
of Transportation
Federal Aviation
Administration

MAJOR REPAIR AND ALTERATION
(Airframe, Powerplant, Propeller, or Appliance)

Form Approved
OMB No. 2120-0020

For FAA Use Only

Office Identification

WPB HLR

INSTRUCTIONS: Print or type all entries. See FAR 43.9, FAR 43 Appendix B, and AC 43.9-1 (or subsequent revision thereof) for instructions and disposition of this form. This report is required by law (49 U.S.C. 1421). Failure to report can result in civil penalty not to exceed \$1,000 for each such violation (Section 901 Federal Aviation Act of 1958).

1. Aircraft	Make Republic	Model RC-3 Seabee
	Serial No. 462	Nationality and Registration Mark N6255k
2. Owner	Name (As shown on registration certificate) Robert A. Gould	Address (As shown on registration certificate) 44-365 Kaneohe Bay Drive Kaneohe, HI 96744-2664

3. For FAA Use Only

4. Unit Identification				5. Type	
Unit	Make	Model	Serial No.	Repair	Alteration
AIRFRAME	(As described in Item 1 above)				X
POWERPLANT					
PROPELLER					
APPLIANCE	Type				
	Manufacturer				

6. Conformity Statement

A. Agency's Name and Address Denis Balczeniak 322 Aolaa Street #1708 Kailua, HI 96734	B. Kind of Agency <input checked="" type="checkbox"/> U.S. Certificated Mechanic <input type="checkbox"/> Foreign Certificated Mechanic <input type="checkbox"/> Certificated Repair Station <input type="checkbox"/> Manufacturer	C. Certificate No. A&P 332400733
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D. I certify that the repair and/or alteration made to the unit(s) identified in item 4 above and described on the reverse or attachments hereto have been made in accordance with the requirements of Part 43 of the U.S. Federal Aviation Regulations and that the information furnished herein is true and correct to the best of my knowledge.

Date
26 February, 2001

Signature of Authorized Individual

7. Approval for Return To Service

Pursuant to the authority given persons specified below, the unit identified in item 4 was inspected in the manner prescribed by the Administrator of the Federal Aviation Administration and is ☒ APPROVED ☐ REJECTED

BY	FAA Fit. Standards Inspector	Manufacturer	<input checked="" type="checkbox"/> Inspection Authorization	Other (Specify)
	FAA Designee	Repair Station	Person Approved by Transport Canada Airworthiness Group	
Date of Approval or Rejection 26 February, 2001		Certificate or Designation No. AI 332400733	Signature of Authorized Individual 	

NOTICE

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8. Description of Work Accomplished

(If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.)

Republic RC-3 Seabee N6255K S/N 462, Time in service 769.2 February 26, 2001

1. Installed Lake & Air, Inc. Amphibian Landing Gear Position Advisory System P/N 9600-1A, S/N 1129 in accordance with STC SA39CH. See drawing #1 for location of indicator lights. See drawing #2 for mounting location of processor unit.
3. Electrical load analysis per preliminary load analysis, reference Figure 1. Ammeter marked and placarded for maximum load per AC 43.13-1B, Chapter 11, Section 3, para 11-36.
4. Circuit breaker installed in circuit breaker panel and wired in accordance with manufacturer's installation instructions and standard practices AC 43.13-2A Chapter 2 para 27. Reference wiring diagram, drawing 3 for detail. Circuit breaker labeled "Gear Warning."
5. Weight and balance data to be incorporated into equipment list. Aircraft to be weighed before return to service.
6. Functional checked and calibrated per Lake & Air, Inc. Amphibian Landing Gear Position Advisory System Installation Manual dated 1-5-94. The installation has been inspected and found to not adversely affect the airworthiness of the aircraft.

Instructions for Continued Airworthiness.

1. Interval: Every annual inspection.
2. Scope: Visually inspect installation for any evidence of deterioration or distress. Check operation per Lake & Air, Inc. Amphibian Landing Gear Position Advisory System Installation Manual dated 1-5-94.

----- Last Item-----

☒ Additional Sheets Are Attached

STC SA39CH

Pages 1,3

Supplemental Type Certificate

Number

SA39CH

This certificate, issued to Lake & Air, Inc.
550 Flying Cloud Drive
Chaska, MN 55318

certifies that the change in the type design for the following product with the limitations and conditions

*therefor as specified hereon meets the airworthiness requirements of Part * of the **

Regulations.

Original Product — Type Certificate Number: *

Make: *

Model: *

*See attached FAA Approved Model List (AML) No. SA39CH for list of approved airplane models and applicable airworthiness regulations.

Description of Type Design Change:

Installation of Amphibian Landing Gear Position Advisory System in accordance with Lake & Air, Inc. Amphibian Landing Gear Position Advisory System Installation Manual as listed on AML No. SA39CH, or later FAA Approved revision.

Limitations and Conditions: 1. Approval of this change in type design applies to the above model aircraft only. This approval should not be extended to aircraft of this model on which other previously approved modifications are incorporated unless it is determined by the installer that the interrelationship between this change and any of those other previously approved modifications, including changes in type design, will introduce no adverse effect upon the airworthiness of that aircraft. (See Page 3 of 3)

This certificate and the supporting data which is the basis for approval shall remain in effect until surrendered, suspended, revoked, or a termination date is otherwise established by the Administrator of the Federal Aviation Administration.

Date of application: March 2, 1993

Date reissued:

Date of issuance: April 21, 1993

Date amended:

January 31, 1994



By direction of the Administrator

Donald P. Michael

Donald P. Michael, Manager
Chicago Aircraft Certification Office

(Title)

Any alteration of this certificate is punishable by a fine of not exceeding \$1,000, or imprisonment not exceeding 3 years, or both.

This certificate may be transferred in accordance with FAR 21.47.

United States of America
Department of Transportation—Federal Aviation Administration
Supplemental Type Certificate
(Continuation Sheet)

Number SA39CH

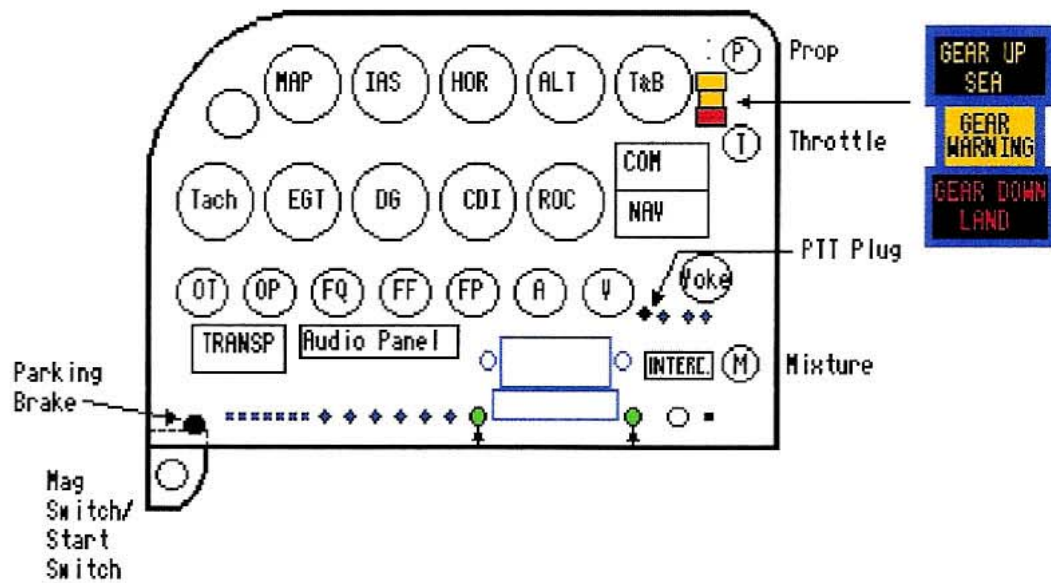
Date Amended: January 31, 1994

LIMITATIONS AND CONDITIONS: (Continued from page 1 of 3)

2. A copy of this certificate and FAA Approved Model List (AML) No. SA39CH amended January 31, 1994, or later FAA Approved revision, must be maintained as part of the permanent records for the modified aircraft.

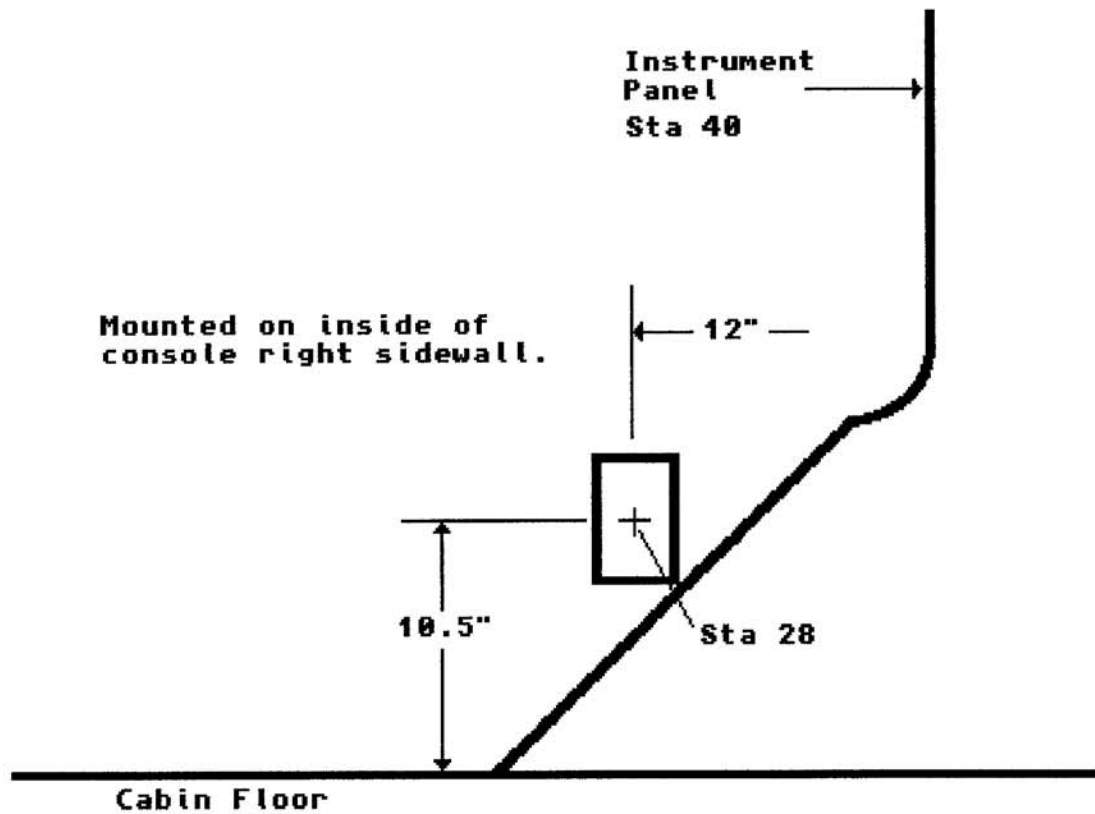
- END -

Drawing #1
Instrument Panel 6255K



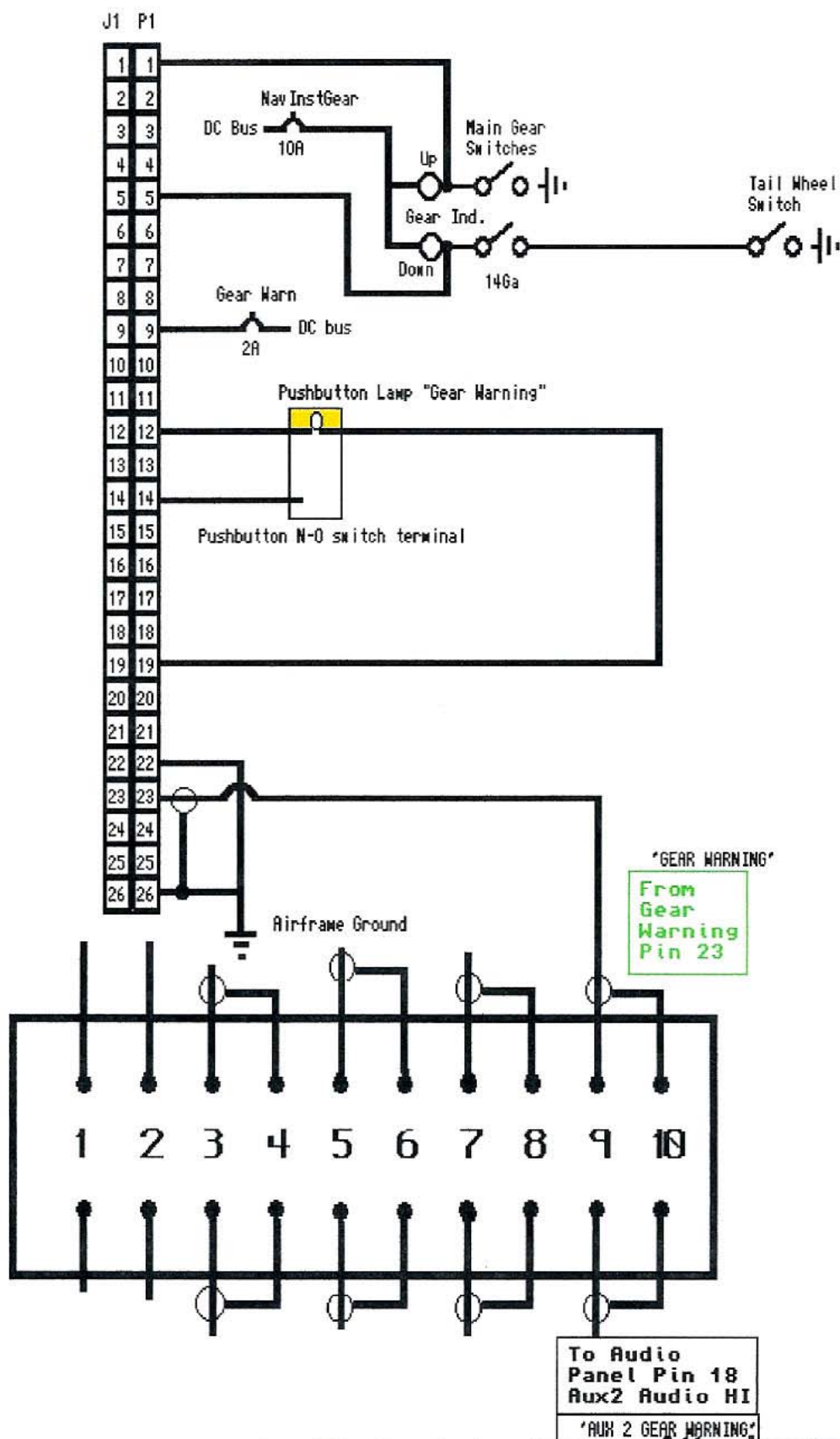
Drawing #2

Showing location of
Gear Position Advisory Processor Unit
Not to Scale



Drawing #3

Processor unit connector



Rev 3/2/01

TS-2
NAV

FIGURE 1

Preliminary Load Analysis

Equipment		Amps
Transponder		0.75
Encoder		0.18
Nav		0.32
CDI		0.60
Com	(Peak Transmit)	2.50
Audio Panel	(Peak)	1.70
Intercom		0.25
EGT Gauge		0.98
Fuel Flow		0.10
Digital Tach		0.20
Gear Warning		0.30
Pulse Light		0.02
Horizon		0.92
DG		0.92
Turn Coordinator		0.50
Instrument Lighting		1.50
Landing Lights	(Intermittent) - (in pulse mode)	7.00
Strobe Lights		7.00
Nav Lights		1.83
Fuel Boost Pump	(Intermittent)	< 5.00
Entertainment System	(Peak)	< 3.00
Engine Instruments		0.20
Bilge Pumps	(Intermittent-total worst case all 3)	< 4.83
Interior light	(Normally not used in flight)	0.67
Anchor light	(Normally not used in flight)	0.61
Total Load, Maximum Possible Intermittent	<	51.88
Normal Total Intermittent Load	<	45.77
Normal Load	<	34.27
Alternator Output, Continuous Rated		50.00
Normal Load as percentage of Alternator output		68.54%
Normal Total Intermittent Load (as percentage of Alternator output)		93.54%
(Inflight, without bilge pumps, interior light, or anchor light)		
Max Intermittent Load as percentage of Alt output		103.76%

Ammeter placarded to indicate max continuous alternator output per AC 43.13-1B, Chapter 11, Section 3, para 11-36. Placard to read:
MONITOR LOAD - DO NOT EXCEED 50 AMPS